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APOLLO SCIENTIFIC EXPERIMENTS DATA HANDBOOK

NASA TECHNICAL MEMORANDUM



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NATIONAL AERONAUTICS AND SPACE ADMINISTRATION

LYNDON B. JOHNSON SPACE CENTER

HOUSTON, TEXAS 77058

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-	Operations Branch W. F. Eichel Update to the Apollo Scientific Experiments Data Handbook	helman man
	The enclosed is an update to the Apollo Scientific Experiments (TMX-58131). This is not a complete reprint, but the changes a accordance with the attached changed pages. Questions about these changes should be directed to Mr. W. W. L JSC Houston, Texas.	Data Handbook are to be made in auderdale, TC3,
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16. Abstract

This publication presents a brief description of each of the Apollo scientific experiments together with the operational history, the data content and formats, and the availability of the data. The lunar surface experiments des ribed are the passive seismic, active seismic, lunar surface magnetometer, solar wind spectrometer, suprathermal ion detector, heat flow, charged particle, cold cathode gage, lunar geology, laser ranging retroreflector, cosmic ray detector, lunar portable magnetometer, traverse gravimeter, soil mechanics, far UV camera (lunar surface), lunar ejecta and meteorites, surface electrical properties, lunar atmospheric composition, lunar surface gravimeter, lunar seismic profiling, neutron flux, and dust detector. The orbital experiments described are the gamma-ray spectrometer, X-ray fluorescence, alpha-particle spectrometer, S-band transponder, mass spectrometer, far UV spectrometer, bistatic radar, IR scanning radiometer, particle shadows, magnetometer, lunar sounder, and laser altimeter. Also included are a brief listing of the mapping products available and information on the sample program.

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Edited by

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29. LUNAR ORBITAL MASS SPECTROMETER (NASA EXPERIMENT S-165)

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29. LUNAR ORBITAL MASS SPECTROMETER

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The lunar orbital mass spectrometer, flown on the Apollo 15 and 16 missions as part of the orbital science payload, measured the concentration of gas molecules it encountered both in lunar orbit and during transearth coast (TEC) for the purpose of studying the sources, sinks, and transport mechanisms of the lunar atmosphere. Nearly 80 hr of operation in lunar orbit and 50 hr in TEC produced some 8000 spectra of gases in the vicinity of the instrument entrance aperture.

INSTRUMENT DESCRIPTION

The instrument, a sector-field, dual-collector mass spectrometer, was mounted on a boom stowed in the scientific instrument module bay of the service module, which was capable of extending the instrument to a distance of 7.3 m from the spacecraft. The purpose of the boom mount was to extend the instrument a reasonable distance from the spacecraft so that it would be beyond the interacting cloud of outgassing molecules from the spacecraft and in a collisionless, outwardly, free-streaming region. The instrument is shown in figure 29-1. The instrument package was a rectangular box, 30 by 32 by 23 cm, weighing 11 kg, and bisected by a baseplate; the electronics portion was on one side, the mass analyzer on the other. A plenum, in the form of a scoop, was mounted on the outboard side of the package and directed along the -X axis of the spacecraft (i.e., opposite the command module end). When the flightpath was oriented so that the velocity vector was in the -X direction, the gas inlet was in the ram direction with respect to native gases in the lunar atmosphere, whereas the reverse direction of flight (+X) produced a wake condition at the inlet.

The plenum contains the mass spectrometer ion source with redundant tungsten (with 1 percent rhenium) filaments mounted on either side of the ionization chamber. Ions formed by electron bombardment are collimated into a beam and accelerated into the mass analyzer, a single focusing permanent magnet, giving a mass resolution of more than a 1-percent valley between peaks at 40 and 41 amu. Two collector systems permit simultaneous scanning of two mass ranges, 12 to 28 amu and 28 to 66 amu.

Voltage scan is employed by using a stepping highvoltage power supply. The ion accelerating voltage sweep is generated by varying the sweep high voltage in a series of 590 steps from 620 V to 1560 V with a dwell time of 0.1 sec/step. Between each sweep, 30 additional steps at zero V are used to determine the background-counting rate and to apply an internal calibration frequency. Therefore, the entire spectrum is obtained every 62 sec, giving a spatial resolution of each mass peak of approximately 100 km. A sweep-start flag indicates data or background and serves as a marker for the start of each sweep. The mass number of the ion being detected is determined by the voltage step number at which the peak is detected. This step number is advanced by an enable pulse from the data-handling system. The minimum number of steps between adjacent mass peaks below mass 54 is 12.

Flectron multipliers, preamplifiers, and discriminators, which count the number of ions that pass through each collector slit on each of the sweep voltage steps, are used in the detector systems. The ion-count numbers are stored in 21-bit accumulators (one for each channel) until sampled by the data-handling system. Just before sampling, each data word is converted to a floatingpoint number in base 2, reducing the data to a 10-bit word consisting of a 6-bit number and a 4-bit multiplier. The data-handling system maintains 7-bit accuracy throughout the 21-bit range of data counts.

Instrument parameters, such as certain internal voltages, electron emission in the ion source, filament currents (to determine which filament is operating), multiplier voltages, sweep voltages, temperatures, multiplier and discriminator settings, and instrument current, are monitored by a housekeeping circuit. The instrument parameters are as follows:

Mass range:	12 to 66 amu
Spectrum scan time:	62 sec
Spatial resolution:	100 km
Mass resolution:	1 percent valley at 40 am
Sensitivity:	10-11 N/m ² (10-13 torr)
Pvnamic range:	10-11 to 10-6 N/m ² (10-13 to 10-8 torr)

Initial calibration of the mass spectrometers, performed in a high-vacuum chamber at the University of

Texas at Dallas, verified that the proper mass ranges were scanned and tested the resolution, linearity, mass discrimination, and dynamic range of the analyzer. Neon was introduced into the vacuum chamber by using isotopic partial pressures ranging from 10^{-9} to 10^{-5} N/m² (10^{-11} to 10^{-7} torr). The instrument response was linear up to 10^{-6} N/m² (10^{-8} torr) where the onset of saturation of the datacounting system occurred. The sensitivity of the instrument was verified to be greater than 2 X 10^{-7} A/N/m² (3 X 10^{-5} A/torr), enabling the instrument to measure partial pressures down to 10^{-11} N/m² (10^{-13} torr). The final absolute calibration was made at the NASA Langley Research Center Molecular Beam Facility in the same manner as for the lunar atmospheric composition experiment (S-205) also described in this handbook.

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OPERATIONAL HISTOPY

The operational history of the Apollo 15 lunar orbital mass spectrometer is given in the following tabulation. Time is in terms of ground elapsed time (GFT) and is measured in hours from lift-off at 13:34:00.79 G.m.t. on July 26, 1971.

<u>Time, hr:min</u>	<u>Spacecraft_attitude</u>
85:05 to 95:15	- X
108:55 to 119:20	- X
130:30 to 141:00	- X
195:50 to 200:25	+ X
202:00 to 211:40	- X
211:40 to 214:15	+ X
224:30 to 238:00	TEC
245:50 to 288:05	TEC

A similar tabulation of the operation of the Apollo 16 experiment follows. The GET is measured from lift-off at 17:54:00.57 G.m.t. on April 16, 1972.

29-5

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<u>t_attitude</u>	Spacecraf	<u>:min</u>	<u>h</u> ;	<u>Time</u> ,
	- X	92:03	to	81:10
	- x	121:20	to	108:00
photographs)	+X (oblique	124:40	to	121:20
	+ X	131:10	to	124:40
	-x	142:10	to	131:10
old	antisolar ho	143:15	to	142:10
	-x	144:15	to	143:15
photographs)	+X (oblique	146:20	to	144:15
	+ X	151:20	to	146:20
photographs)	+X (obligue	152:50	to	151:20
	* X	164:20	to	152:50
	+ X	167:00	to	164:20
photographs)	+X (oblique	168:10	to	167:00
	- X	193:45	to	180:05

FORMAT OF DATA

Nata processing has resulted in the blocking of data into complete mass spectra on magnetic tape. Brief time gaps in the data, caused by telemetry dropouts, are filled with flag words to ensure proper location of the good measurements in the spectra. Reduced data also include the background-count level of each analyzer channel, the amplitude of each mass peak, the decommutated housekeeping data, and the pertinent spacecraft-trajectory information.

Microfilm records are formatted outputs of the data on magnetic tape. The format gives sequential pairs of mass spectra (high- and low-mass channels) together with background, peak amplitude, housekeeping, and trajectory data. Periodic tabulated summaries of the peak amplitudes, housekeeping, and trajectory data are also given. Each summary covers several hours of experiment operation.

SAMPLES OF DATA

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Computer printouts showing examples of each type of data are given in figures 29-2 to 29-6. The codes and definitions are as follows:

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Low Mass

GET TIME	GET from lift-off
APOLLO 16	Apollo mission number
AO, A1, BO, B1	Peak-location coefficients for high-mass channel
₽ Ŧ V	Orbit number
SUN HR	Longitude – subsolar longitude
LONG	Subsatellite longitude
LAT	Subsatellite latitude
SUN A	Angle from commend and service module (CSM) x axis to Sun
SUN B	Azimuth of Sun from -Z axis
TEMP	Ion-source temperature monitor
ALT	CSM altitude (kilometers)
VEL ALPHA	Angle of attack from X axis
VFL	CSM velocity (km/sec)
BACKGROUND	Background data
CALIBRATE	Interval calibrate data
HOUSEKEPPING	
+12	+12 V monitor
+ 5	+5 V monitor
-12	-12 V monitor
-15	-15 V monitor

29-7

-15 V monitor

EM	Emission current monitor
P 1	Filament 1 current monitor
F2	Filament 2 current monitor
LM	Low-mass multiplier, high-voltage monitor
HM	High-mass multiplier, high-voltage monitor
SW	Sweep high-voltage monitor
T1	Electronics temperature monitor
Τ2	Ion-source temperature monitor
MF	Multiplier high-voltage HI/LO flag
DF	Discriminator HI/LO flag
IC	Instrument current
PEAK AMPLITUDES	Mass number and peak amplitude
PRE BKG	Background counts at start of sweep
CUR BKG	Background counts at end of sweep
NUM	Number data points used in calculating BKG
	High Mass
GET TIME	GET from lift-off
APOLLO 16	Apollo mission number
24 October 1972	Date data tapes were processed
AO, A1, BO, B1	Peak-location coefficients for high mass channel
STEP NO	AO + A1/mass step <400 BO + B1/mass step >400

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SYNC START Data count at start of sweep SYNC END Data count at end of sweep PBKG Background count at start of sweep BKG Background count at end of sweep Number data points used in N calculating BKG BACKGROUND Background data Internal calibrate data CALIBRATE PEAK AMPLITUDES Mass number and peak amplitude STAR Incorrect peak shape Apollo 16 Peak Summary GET TIME GET from lift-off SUN HR longitude - subsolar longitude С Peak-error code. If code greater than zero, previous peak-location coefficients are used Mass number L Low-mass BKG H High-mass BKG GAP Time gap in data STAR Incorrect peak shape CSM DIFECTION Minus (-) denotes -X orientation

Apollo 16 Trajectory Summary

GET TIME	GET	from	lift-off
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REV	Orbit number
SUN HR	Longitude – subsolar longitude
LONG	Subsatellite longitude
LAT	Subsatellite latitude
PADIUS	Orbit radius (kilometers)
VELOCITY	CSM velocitv (km/sec)
ALTITUDE	CSM altitude (kilometers)
S LONG	Subsolar longitude
SJ LAT	Subsolar latitude
SUN A	Angle from CSM X axis to Sun
SUN B	Aziwuth of Sun from -Z axis
VEL ALPHA	Angle of attack from X axis
VEL BETA	Azimuta of Vel vector from -Z axis

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Apollo 16 Housekeeping Summary

GET TIME	GET from lift-off
+12	+12 V monitor
+5	+5 V monitor
-12	-12 V monitor
-15	-15 V monitor
FMISSION	Emission current monitor
FII, 1	Filament 1 current monitor
FIL 2	Filament 2 current monitor
TW HV	Low-mass multiplier, high-voltage monitor
HN HV	High-Mass multiplier, high-voltage morttor

S HV	Sweep high-voltage monitor
ETEMP	Electronics temperature monitor
S ፹ ፑ ዛ ዖ	Ion source temporature monitor
M 1.0/HI	Multiplier high-voltage 10/HI flag
P HI/LO	Discriminator HI/LO flag
I	Instrument current (total)

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Figure 29-2.- Low-mass data format.

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Figure 29-3.- High-mass data format.

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	VEL ALPHA	177.00	176.65	176.47	175.45	119.29	175.44	175.56	175.60	175.62	175.59	175.52	175.46	176.06	176.92	176.92	110.47	1/0.1	10.011	174.00	110.01	111.01	177 71	177 98	178.17	178.12	178.42	170.46	170.37	170.16	177.06	11.50	171 70	176.29	176.22	176.42	176.55	176.63	19.9/1	10.71	178.01	176.49	117.57	177.00	80.771	177.16	
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1 OMC	91.49	(C. 9)	12.22	b9.14	66.06	62.96		52.75	50.68	17.60	44.52	41.44	30.35	35.26	32.17	29.07	25.97	22.87	17.76	16.67	14.21	14.01	92.7		20.1	11.2-		-11.54	-14.69	-17.03	-20.98	-24.13	82.12-	21-06-		- 10	-42.99	-46.13	-49.26	-52.39	-55.51	0.00		-61.97	-71.00	-79.16	
SIN ND	33.10	20.02	23.86	20.79	17.72	14.65	611.00 61 50	5.45	2.38	69	-3.76	-6.84	16.6-	-12.99	-16.03	-19.16	cz. 27-	-22-32								97.04	19.95-	-59.66	-62.80	-65.94	-69.08	-72.22	-/- 16		70.40	-87.90	-91.03	-94.16	-97.20	100.40	201.	Co. 001	-112.64	-115.94	-119.03	71.221.	
RFV	58.27	58.29	56.30	50.31	56.32	20.55	58.34	50.35	50.35	50.37	50.30	50.30	58.39	58.40	50.41	24.90											58.52	58.53	58.54	59.55	58.56	58.57	70.70	55 50	50.60	56.61	50.62	50.63	58.64			58.67	50.68 -	56.59 -	28.20		
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Figure 29-5.- Trajectory summary format.

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AN CA			10.7	19.7	2.67	2.69	2.67	2.67	2.71	2.69	2.69	07 0		10.7	2.11	2.69	2.69	07 0		20.2	2.69	2.71	2 49		7	2.69	2.71		1	2.71	2.69	1		2.11	2.71	2.71	2.71	11.0		1	Z.69	2.69	2.69	2.69	09.6			10.2	2.67	2.69	2.69	2.69	2.69	2.69	2.69	2.67	2.69	2.69	2.71	
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APOLLO Enission	2.06	200	5	200	50	32	5.0	5.0	90.2	2.06	2.05	2.10	10			2.04	2.04	2,05	00	54	90.7	2.04	2.05	40.0	5	50.7	2.05	000			2.02	20.0	200	5.0	5.5	2.C	2.04	2,04		200	20.2	z. 05	1.96	2.02	2.00	2.00	5	20.40	5	20.0	2.04	2.02	2.04	2.04	2.04	2.02	2.02	2.04	2.05	
-15	4.67	4 47	4 47	1.7.1						4.67	4.67	4.67	4.67	1 1		-0.	4.67	4.69	4.47			19.4	4.67	4.67		1.67	4.65	1 47		19.1	4.67	4.67				4.67	4.67	4.67	1 47			1.6/	4.65	4.67	4.67	4.67	1 4 7	10.4			19.4	4.67	4.67	4.67	4.65	4.67	4.67	4.65	4.65	
-12	2.96	2.95	2.96						14.7	2.46	2.96	2.96	2.96	20 6	2	2.76	2.95	2.96	2.96			6.76	2.96	2.96		4.70	2.94	20 4		2.7	2.54	2.96				2.46	2.96	2.96	20 6		0 · · 70	2.74	2.96	2.96	2.96	2.96	20 6			0	2.76	2.94	2.96	2.96	2.96	2.94	2.96	2.96	2.96	
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+12	3.08	3.03	52.6	1.86					33	20 10	3.06	3.05	3. 83	20 4		5.	3.06	3.03	3.83			20.7	3.06	3.85		1 0.1	3.56	2 20		2.00	3.06	3.03			00.0	J. UJ	3.06	3.86	1.06			1.00	3.85	3.85	3.88	3.66	1.04				20.1	J. 03	3.86	3. 88	3.86	3.88	3.85	3.88	3.86	
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Figure 29-6.- Housekeeping summary format.



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(a) Near side.

Figure B-6.- Lunal surface groundtrack envelope of the Apollo 15 orbiting spacecraft for revolutions 1 to 74. Areas of additional data coverage outside the envelope are determined by the fields of view of experiment instruments and photographic cameras.



(b) Far side.

Figure B-6.- Concluded.

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(a) Near side.

Figure B-7.- Lunar surface groundtrack envelope of the Apollo 16 orbiting spacecraft for revolutions 1 to 65. Areas of additional data coverage outside the envelope are determined by the fields of view of experiment instruments and photographic cameras.

B-8





B-9



APPENDIX C

LIST OF ACRONYMS

A DC analog-to-digital converter AET Apollo elapsed time AGRS Apollo gamma-ray spectrometer AIC Apollo intermediate charts ALSE Apollo lunar sounder experiment ALSEP Apollo lunar surface experiments package ASE active seismic experiment ATS Applied Technology Satellite AXRS Apollo X-ray spectrometer λZ azimuth BCD binary coded decimal cold cathode gage experiment CCGE CCIG cold cathode ion gage CDC Control Data Corporation CDR commander center of gravity c. g. charged-particle lunar environment experiment CPLEE CRT cathode ray tube CSAR coherent synthetic aperture radar CS M command and service module CTE computer time elapsed Cyq Cygnus declination dec DRNM Deep River Neutron Monitor DSN Deep Space Network DTREM dust thermal radiation engineering measurement early Apollo scientific experiments package EASEP E-frame electronic frame EI, elevation EMI electromagnetic interference EOF end of file EOT end of tape FΡ explosive package Environmental Research Institute of Michigan ERIM EVA extravehicular activity FET field effect transistor FFT fast Fourier transform FM frequency modulation FOV field of view FWHM Eull width, half maximum GCR galactic cosmic ray GE General Plectric GET ground elapsed time

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G.m.t. Greenwich mean time GRS gamma-ray spectrometer GS FC Goddard Space Flight Center high frequency HF HFE heat flow experiment ΗV high voltage TMP Interplanetary Monitoring Platform TR infrared ISR infrared scanning radiometer JPL Jet Propulsion Laboratory JSC Lyndon B. Johnson Space Center LAC lunar astronautical charts lunar atmospheric composition experiment LACE Lamont-Doherty Geological Observatory L-DGO LEAM lunar ejecta and meteorites LLT local lunar time lunar module LM large Magellanic Cloud LNC lunar neutron probe experiment LNPE long period LP lunar portable magnetometer LP 🗄 LPX long period horizontal (X-axis) or long-period horizontal seismometer LPY long period horizontal (Y-axis) or long-period horizontal seismometer long period vertical (7-axis) or long-period LPZ vertical seiscometer LRC Langley Pesearch Center lunar roving vehicle LRV Lunar Sample Analysis Planning Team LSAPT lunar surface gravimeter LSG LSI Lunar Science Institute LSM lunar surface magnetometer LS PE lunar seismic profiling experiment LURE Lunar Laser Ranging Experiment low-voltage power supply LV PS MA mass analyzer modularized equipment stowage assembly MESA modularized equipment transporter MET MPA mortar package assembly memory readout MRO NASA Apollo trajectory NAT National Bureau of Standards NBS National Space Science Data Center NSSDC Orbiting Astronomical Observatory 2 010-2 OGO IV Orbiting Geophysical Observatory IV PA post amplifier PCM pulse code modulation PFS particles and fields subsatellite PMT photomultiplier tube PSD pulse shape discriminator passive seismic experiment PSE

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PSEP	passive seismic experiments package
RA	right ascension
rev	revolution
RFI	radiofrequency interferometry
RLC	Ranger VII lunar charts
rms	root mean square
RT	real time
RTG	radioisotope thermoelectric generator
SA	SIM attitude
SAO	Smithsonian Astrophysical Observatory
SB	standby
Sco	Scorpius
SE	solar ecliptic
SEM	scanning electron microscope
SEP	surface electrical properties
SIDE	suprathermal ion letector experiment
SIM	scientific instrument module
SIVB	Saturn IVB
SM	solar magnetospheric
SP	short period
SRI	Stanford Research Institute
SRP	self-recording penetrometer
SWS	solar-wind spectrometer
TCE	telemetry conversion error
TEC	transearth coast
TEI	transearth injection
TGE	traverse gravimeter experiment
TID	total ion detector
TSF	telemetrv-store fast
TSN	telemetry-store normal
UCLA	University of California at Los Angeles
USGS	U.S. Geological Survey
OTD	University of Texas at Dallas
UV	ultraviolet
UVS	ultraviolet spectrometer
VCO	voltage-controlled oscillator
VHF	very high frequency
VLBI	very long paseline interferometry
VSA	vibrating string accelerometer
WUCHAHRUS	world para Center A for Pockets and Satellites

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